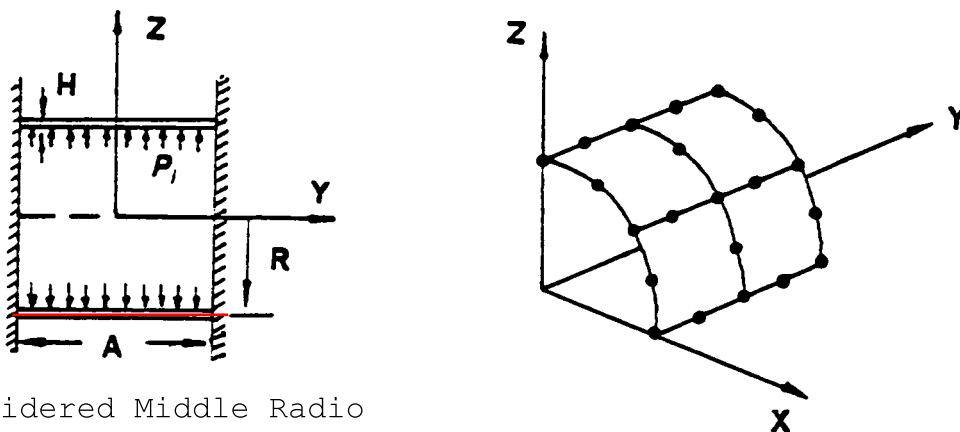


## 6.11 PROBLEM 10 : COMPOSITE CYLINDRICAL SHELL

The geometry, loading and finite element mesh of a clamped cylindrical shell subjected to internal pressure is shown below:



Considered Middle Radio

$$\begin{array}{ccc}
 \frac{1}{E_1} & \frac{-v_{21}}{E_2} & \frac{-v_{31}}{E_3} \\
 \frac{-v_{12}}{E_1} & \frac{1}{E_2} & \frac{-v_{32}}{E_3} \\
 \frac{-v_{13}}{E_1} & \frac{-v_{23}}{E_2} & \frac{1}{E_3}
 \end{array}
 \quad \text{Modern Convention}$$

$$R = 20 \text{ in.}, A = 20 \text{ in.}, H = 1 \text{ in.}, P_i = 2.0403664 \text{ psi}$$

The lamina properties are :

$$E_1 = 7.5 \times 10^6 \text{ psi}, \quad E_2 = 2 \times 10^6 \text{ psi},$$

$$G_{12} = 1.25 \times 10^6 \text{ psi}, \quad G_{12} = G_{13} = G_{23}, \quad v_{12} = v_{13} = 0.25$$

NUMERICAL RESULTS

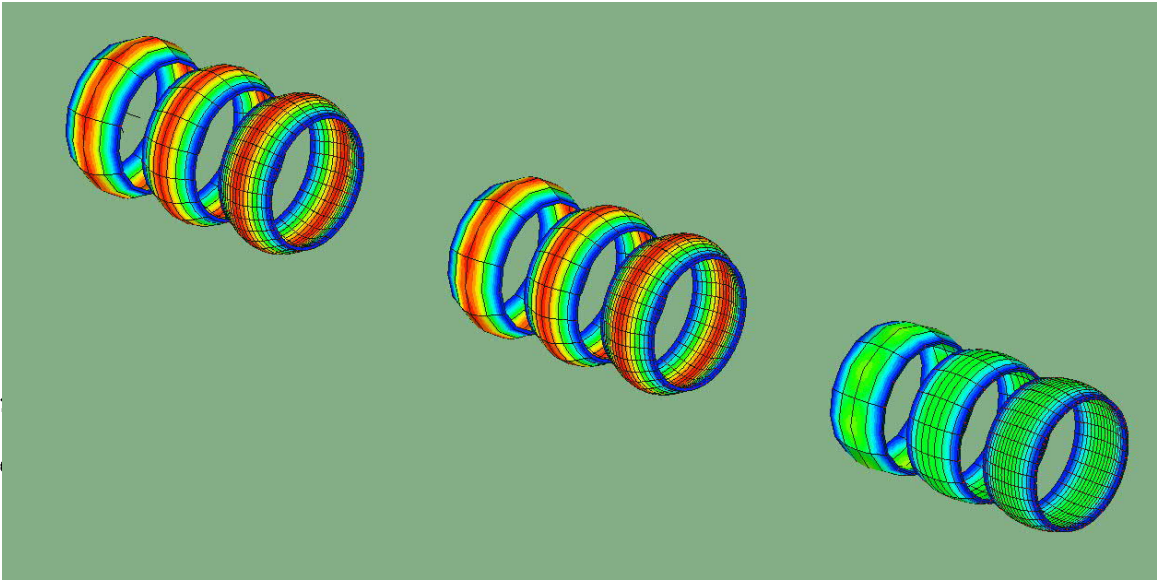
This model is used for both static and transient analysis. In Table 6.19, the center deflections for orthotropic and two-layer (0/90) cross-ply shells are compared with known solutions.

**Table 6.19 Comparison of the Center Deflection (Inch)**

Lamination Static	Present	Ref.[1]	Ref.[72]	Analytical [73]	ccx	ccx	Internal	Internal
Schema	Work 2x2Q8	2x2Q9			2x4	4x8	2x4	4x8
Cylindrical_Shell_Orthotropic S8R					0,000369	0,000362	0,000304	0,000360
Cylindrical_Shell_1_Ply_0	0,0003706	0,0003727	0,0003666	0,000367	0,000369	0,000362	0,000304	0,000360
Cylindrical_Shell_Cross_Ply_0/90	0,0001841	0,0001803			(0,0001607 - 0,0002488)	0,0001857	(0,000170 - 0,000209)	0,000183

Units w-[inch]

Present Work refers to : **LARGE DEFORMATION ANALYSIS OF LAMINATED COMPOSITE STRUCTURES BY A CONTINUUM-BASED SHELL ELEMENT WITH TRANSVERSE DEFORMATION . Pey M. Wung .(1989)**



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 73. Timoshenko, S. and Woinowsky-Krieger, S., *Theory of Plates and Shells*, 2nd Ed., 1959. -

74. Chandrashekhara, K., "Geometric and material nonlinear analysis of laminated composite and shells," Ph.D. thesis, VPI&SU, Blacksburg, VA, 1985.